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RESEARCH MEMORANDUM

WIND-TUNNEL INVESTIGATION AT SUBSONIC AND SUPERSONIC SPEEDS OF A MODEL OF A TAILLESS FIGHTER AIRPLANE EMPLOYING A LOW-ASPECT-RATIO SWEPT-BACK WING - EFFECTS OF EXTERNAL FUEL TANKS AND ROCKET PACKETS ON THE DRAG CHARACTERISTICS

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RESEARCH MEMORANDUM

WIND-TUNNEL INVESTIGATION AT SUBSONIC AND SUPERSONIC SPEEDS OF A MODEL OF A TAILLESS FIGHTER AIRPLANE EMPLOYING A LOW-ASPECT-RATIO SWEPT-BACK WING - EFFECTS OF EXTERNAL FUEL TANKS AND ROCKET PACKETS ON THE DRAG CHARACTERISTICS

By Willard G. Smith

SUMMARY

The effects of external fuel tanks and externally mounted rocket packets on the drag characteristics of a model of a tailless fighter airplane are presented in this report. The investigation was conducted through a Mach number range of 0.60 to 0.90 and 1.20 to 1.70 at a constant Reynolds number of 3.2 million. The measured lift, drag, pitching-moment, and rolling-moment coefficients and lift-drag ratios are presented in tabular form and the drag characteristics and lift-drag ratios are also presented in graphic form. In addition, pressure distribution data are tabulated which may be used to determine the influence of the external stores on the wing load distribution at supersonic speeds.

Results of this investigation show that the addition of two external fuel tanks and four faired rocket packets to the model produced drag increments which increased from 30 percent to 50 percent of the drag of the basic model between Mach numbers of 0.60 and 0.90, respectively, while at supersonic Mach numbers this drag increment was approximately 30 percent of the drag of the basic model. Tests of the model fitted with four rocket packets indicate that the drag may be reduced at subsonic speeds by fairing the open rocket packets, but at supersonic speeds the faired packets produced more drag. A small decrease in drag was realized at supersonic speeds, for the model fitted with two fuel tanks and four rocket packets, by mounting the outboard packets and fuel tanks in a more forward chordwise position with respect to the wing.

INTRODUCTION

Knowledge of the increases in drag to be expected from the addition of externally mounted fuel tanks and armament under the wings and fuselage becomes increasingly important as the trend continues toward long-range, high-speed fighter airplanes carrying rocket-propelled armament. An



investigation of the effects of this type of external installation on the aerodynamic characteristics of a model having a low-aspect-ratio swept-back wing has been conducted in the Ames 6- by 6-foot supersonic wind tunnel. The model was fitted with various combinations of underthe-wing type rocket-packet and fuel-tank installations and tested at subsonic and supersonic Mach numbers at a constant Reynolds number. Two chordwise locations of the fuel tanks and rocket packets were investigated and the rocket packets were tested with the ends of the packets faired smooth and with the rocket tubes open. The results of this investigation are presented herein. The results of an investigation of the stability and control characteristics of this same model conducted in the Ames 6- by 6-foot supersonic wind tunnel are presented in reference 1.

NOTATION

The lift, drag, and pitching-moment coefficients are referred to the stability axes with the origin at the quarter-chord point of the mean aerodynamic chord projected to the fuselage center line. Rolling-moment coefficients are referred to the fuselage longitudinal axis.

р	wing span, feet
c	local wing chord measured parallel to plane of symmetry, feet
ē	wing mean aerodynamic chord $\left(\frac{\int_{0}^{b/2} c^{2} dy}{\int_{0}^{b/2} c^{2} dy}\right)$, feet drag coefficient $\left(\frac{drag}{qS}\right)$
$c_{\mathbf{D}}$	$\operatorname{drag \ coefficient}\left(\frac{\operatorname{drag}}{\operatorname{qS}}\right)$
$c^{D^{\mathbf{B}}}$	increment of drag coefficient due to external-store installation or fuselage modification based on total wing area $(c_{D_{model}} + store^{-C_{D_{model}}})$
$\mathtt{C}_{\mathbf{L}}$	lift coefficient $\left(\frac{\text{lift}}{\text{qS}}\right)$
Cl	rolling-moment coefficient $\left(\frac{\text{rolling moment}}{\text{qSb}}\right)$
$C_{\mathbf{m}}$	pitching-moment coefficient $\left(\frac{\text{pitching moment}}{\text{qS}\overline{c}}\right)$
$\mathtt{c}_{\mathtt{p}}$	static pressure coefficient $\left(\frac{p-p_0}{q}\right)$
$\frac{\mathbf{L}}{\mathbf{L}}$	lift-drag ratio



$\left(\frac{\overline{D}}{D}\right)_{\max}$	maximum lift-drag ratio
M	free-stream Mach number
p	local static pressure, pounds per square foot
p_{o}	free-stream static pressure, pounds per square foot
q	free-stream dynamic pressure, pounds per square foot
R	Reynolds number, based on the mean aerodynamic chord
s	total projected wing area, including area formed by extending leading and trailing edges to plane of symmetry, square feet
Y	spanwise distance from plane of symmetry, feet
α	angle of attack of fuselage longitudinal axis, degrees

APPARATUS

Wind Tunnel and Equipment

The present investigation was conducted in the Ames 6- by 6-foot supersonic wind tunnel. This is a closed-return, variable-pressure wind tunnel in which the pressure and Mach number can be continuously varied. The stagnation pressure can be varied from 2 to 17 pounds per square inch absolute and the Mach number can be varied from 0.60 to 0.90 and from 1.15 to 2.00. A complete description of the wind tunnel is given in reference 2.

The model was sting mounted with the pitch plane of the model horizontal in the wind tunnel to utilize the most uniform stream conditions. (See reference 2). A four-component electrical strain-gage balance, similar in design to that used in reference 3, was enclosed within the fuselage of the model. The aerodynamic forces and moments were registered by recording-type galvanometers calibrated by applying known loads to the balance.

Model

A model of a high-speed fighter airplane having a low-aspect-ratio, swept-back wing and a swept-back vertical tail but not horizontal tail was used in this investigation (fig. 1). A bubble-type canopy was faired into a dorsal fin which extended back to the vertical tail. Provisions



were made for fairing the vertical tail into the fuselage when the canopy and dorsal fin were removed. The wing had a leading-edge sweep angle of 52.5° and a taper ratio of 0.332 based on the theoretical wing tip. The wing was composed of symmetrical sections in streamwise planes having a thickness of 7.0 percent of the chord at the wing root tapering to 4.5 percent of the chord at the theoretical wing tip.

The model was fitted with inlets housed in wing-body juncture fairings with internal ducts allowing the air to flow through and exhaust at the rear of the fuselage. In this investigation the mass flow of air through the ducts was not adjustable; however, the ducts were constructed so that at supersonic speeds the exit was choked, limiting the inlet Mach number to 0.4. In order to accommodate the annular duct exit and the mounting sting, the boattailing on the model was somewhat less than would be expected on a full-scale airplane.

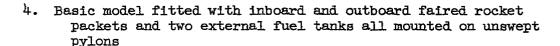
Rocket packets and fuel tanks were provided, to be attached to the wings in the locations shown in figures 2 and 3. The outboard rocket packets and the fuel tanks were mounted on unswept and swept-forward pylons as shown in figures 2 and 3. The purpose of the swept-forward pylons was to obtain a more forward location of these stores. The rocket packets were tested both with the fore and aft ends of the rocket packet faired smooth and with six holes open through the packet, to simulate conditions before and after firing the rockets.

Provisions were made to measure pressure distribution data at five spanwise stations as shown in figure 4. The location of the orifices on the upper and lower surfaces of the port wing are given in table I.

TESTS AND PROCEDURE

As a basis for comparison, tests were made of the basic model with canopy and dorsal fin in place and with no external stores installed. Lift, drag, pitching-moment, and rolling-moment data were obtained at Mach numbers of 0.60, 0.80, 0.90, 1.20, 1.35, 1.50, and 1.70 at a constant Reynolds number of 3.2 million, through an angle of attack range of -2° to +8°. Similar data were then obtained at corresponding test conditions for the following model configurations:

- 1. Basic model fitted with inboard and outboard faired rocket packets mounted on unswept pylons
- 2. Basic model fitted with inboard and outboard open-tube rocket packets mounted on unswept pylons
- 3. Basic model fitted with two external fuel tanks mounted on unswept pylons



- 5. Basic model fitted with outboard faired rocket packets and two external fuel tanks mounted on swept pylons and inboard faired rocket packets mounted on unswept pylons
- 6. Basic model with canopy and dorsal fin removed (no external stores)

Pressure distribution data were obtained for the basic model and for the model fitted with four faired rocket packets mounted on straight pylons. These tests were conducted at Mach numbers of 1.20, 1.30, and 1.70 at a Reynolds number of 2.0 million. Data were obtained through an angle-of-attack range of -3° to $+12^{\circ}$ at 2° increments for the basic model and 4° increments for tests of the model fitted with the rocket packets. A tabulation of the test conditions is presented in table II.

Reduction of Data

The test data have been reduced to standard NACA coefficient form based on the total projected wing area including the area in the region formed by extending the leading and trailing edges to the plane of symmetry (fig. 1). Factors which could affect the accuracy of these results and the corrections applied are discussed in the following paragraphs.

Angle of attack. The determination of the actual angle of attack of the model under load required several corrections to be applied to the nominal angle. Corrections, determined from static load calibrations, were applied for the angular deflection of the sting and balance under aerodynamic load and for the angular movement due to structural clearance in the model support and balance. These corrections amounted to from 5 to 10 percent of the nominal angle, depending on the load.

Tunnel-wall interference. Corrections to the data for the effects of the tunnel walls at subsonic speeds were made by the method of reference 4. These corrections which were added to the data were as follows:

$$\Delta \alpha = 0.377 C_{\rm L}$$

$$\Delta C_D = 0.0066 C_L^2$$

The reflected bow wave did not intersect the model and so no tunnel-wall corrections were made for supersonic Mach numbers.



CONTINUENTAL

The effect of constriction of the flow at subsonic speeds due to the presence of the model was taken into account by the method of reference 5. This correction was calculated for conditons of zero angle of attack and was applied through the angle-of-attack range. At a Mach number of 0.90, this correction amounted to a 1-percent increase in Mach number and dynamic pressure over those values determined from calibrations of the wind tunnel without a model in place.

Support interference. Results of a wind-tunnel test of a similar model (reference 6) show that the effects of support interference consisted primarily of a change of pressure at the base of the model. In this test the base pressure was measured and corrections were applied to adjust the pressure at the base to free-stream static pressure. The drag values are, therefore, forebody drag coefficients.

Stream variations.— Tests were made at subsonic and supersonic speeds with the model in upright and inverted attitudes. Results of these tests showed no measurable indications of stream angle or stream curvature in the horizontal plane of the wind tunnel. Stream surveys of the Ames 6- by 6-foot supersonic wind tunnel (reference 2) show some curvature in the vertical plane of the wind tunnel, but the results of a subsequent investigation (reference 7) indicate that this curvature has little effect on the longitudinal aerodynamic characteristics of the model when pitched in the horizontal plane.

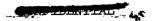
Internal duct drag. - The model was equipped with twin ducts through which air could flow. However, provisions were not made to vary the mass flow, so a study of the duct drag characteristics was not feasible in this investigation. The drag data presented herein are for the complete model; that is, the drag due to flow through the ducts has not been subtracted from the final drag coefficients.

Precision of Data

The accuracy of the test results, excluding stream effects, is shown by the repeatability of the data. Examination of the results showed the data to repeat with the accuracy shown in the following table:

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The base area used in this investigation was the entire base area of the model less the duct exit area.



	Accı	racy
Quantity	$C_{L} = 0$	$C_{L} = 0.25$
$\mathbf{c}_{ ext{D}}$	±0.0004	±0.0006
$c_{ m L}$	±.0016	±.0018
C_{m}^{-}	±.0005	±.0005
Cl	±.0006	±.0009
$c_{f p}$	±.005	±.005
Μ	±.03	± •03
R	±.03 × 10	± .03 × 10 ⁶
α.	±.1	±.15

The precision of the data presented herein is superior to that of the data in reference 1 because these data were obtained for a consecutive series of tests in the wind tunnel and the mounting of the model and balance was unchanged during this investigation.

RESULTS AND DISCUSSION

Only the data pertinent to a study of the effects of external fuel tanks and rocket packets on the drag characteristics of the model are discussed in this report. All the force and moment data obtained from these tests, including lift and rolling-moment coefficients and lift-drag ratios, are presented in table III, however. In addition, experimental static pressure coefficients obtained at Mach numbers of 1.20, 1.30, and 1.70 for the basic model and for the model fitted with four rocket packets are presented in table IV. Comparison of the data from these pressure distribution tests gives an indication of the effects of the rocket-packet installation on the air loads experienced by the model.

The effects of external stores on the drag characteristics of the model are presented in this report as the increments of drag coefficient incurred by the addition of external stores. Figure 5 presents the variation of drag coefficient with lift coefficient for the basic model at Mach numbers of 0.60, 0.80, 0.90, 1.20, 1.35, 1.50, and 1.70. As previously mentioned, the drag coefficients presented in this report include the internal duct drag. The increments of drag coefficient for the various store installations investigated are shown in figure 6 as a function of Mach number for 0 and 0.25 lift coefficients. This figure shows that at subsonic speeds the drag increment resulting from the addition of four rocket packets was somewhat less when the packets were faired, but at supersonic speeds fairing the packets increased the drag. The drag increments for two fuel tanks and four rocket packets, mounted in the aft chordwise location (unswept pylons), varied from approximately 30 percent of the drag of the basic model at a Mach number of 0.60 to 50 percent at

a Mach number of 0.90. For Mach numbers of 1.20 to 1.70 the drag increment for these same external-store configurations was approximately 30 percent of the drag of the basic model. Results of tests of the model with the stores mounted in two chordwise locations showed that the change in chordwise location had no significant effect on the drag at subsonic speeds. At supersonic speeds, however, the drag increment resulting from the addition of two fuel tanks and four rocket packets was somewhat smaller for the forward chordwise location (swept pylons).

The maximum lift-drag ratios for all the configurations tested are shown in figure 7 as a function of Mach number. These data are for the unbalanced model.

Results of this investigation show that the addition of external stores could appreciably affect the trim drag of the model. This effect is illustrated in figure 8 which shows the variation of pitching-moment coefficient with lift coefficient for the basic model and for the model fitted with two external fuel tanks and four rocket packets. The magnitude of the pitching-moment coefficient at zero lift for the basic model was quite small at all Mach numbers, but the model fitted with external stores showed a significant negative pitching moment at subsonic speeds and a positive pitching moment at supersonic speeds. These pitching moments, associated with the installation of external stores on the model, significantly influence the deflection of the longitudinal control surface required for a specific flight condition. Thus it should be noted that the drag coefficients presented for this investigation are for the unbalanced model and that the total drag for the model balanced with a control device will include an additional drag increment or decrement due to the change in control setting required to counteract the aerodynamic influence of the external store. Pitching-moment characteristics are shown for the model fitted with two fuel tanks and four rocket packets because they exhibit the most pronounced effects of external stores of all the configurations investigated.

CONCLUSIONS

The following conclusions are based on a wind-tunnel investigation of the effects of external fuel tanks and externally mounted rocket packets on the drag characteristics of a model of a tailless fighter airplane:

1. The drag increase resulting from the addition of two external fuel tanks and four faired rocket packets varied from 30 percent of the drag of the basic model at 0.60 Mach number to 50 percent of the drag of the basic model at 0.90 Mach number. At Mach numbers of 1.20 to 1.70, this drag increment was approximately 30 percent of the drag of the basic model.

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- 2. The drag coefficient, at subsonic speeds, for the model fitted with four faired rocket packets was smaller than with four open rocket packets. At supersonic speeds the four faired packets produced greater drag increments than the open packets.
- 3. The drag coefficients for the model fitted with two fuel tanks and four faired rocket packets were somewhat less, at supersonic speeds, with the outboard rocket packets and fuel tanks in a forward chordwise location. At subsonic speeds the chordwise location caused no significant effect on the drag characteristics.

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National Advisory Committee for Aeronautics
Moffett Field, Calif.

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TABLE III.- CONCLUDED
(d) Tests 39 through 48



TABLE IV.- EXPERIMENTAL PRESSURE COEFFICIENTS, C_p (a) Basic model, M = 1.2

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20	- 114	178	- 900	239	268	- 332	388	432	314	77	53	160	137	116		054	.012	.079	.JA6	.209	. <u>ě</u> 79
<u> 20</u>	.024	006	.017	061		131	167	- 207	-261	366	, j	209	186			118		.003	.076	.209 .094 .041	-153
22	094	.044	.022	005	018	073	خند-ا	259	197	008	77	178	175	[15%	123	168	 068	029	.009	.044	.085
23	.069	.026	.001	022	070	094	126	163	199	226	RXXX88					t	la:	i			
24	-053	.010	017	038	076	096	130	169	- 805	229	<u>ग</u>	02	-186						010	-051	-012
25	.077	007	037]06 0	08IL	128	159	193	235	271	_ Z	16 .185	-· <u>:</u>	168		Ի	102 369	<u>or</u> _	633	.005 716	.020
26			} 	- <i></i>					 -	i-	Z 2	.077	05	026 144	124 195		1.73	520 530	L.611	- 639	747 733
27					- 5.5			h ===	1-5	F 5.2	8	.027	068	110	1.37	226	398	二烷	606	679	723
	26+	달	325	330	344	366 065	323	ŀ:数	┝-왔		62	.016	032	092	343		228	드살	F:33	695	F.76
89	424	. 63	쁘	276 366	222 309	234	.065 .088	1.26	.305 .155		69	00*	069			-348	209	1	- 463	763	- 676
30	-1/6	115	110	002	166	04	.005	.036	.007	.26 .12	63 64	- 033	098	126	143	1-175	215	J064	Link	535	- 624
32 I	097	067	046	020	.005	l.öii	.087	139	194	291	65	078	138	172	197	- 201	- 263	290	i 372	500	770
39	111		065		009	.ola	.113	1.178	.233		66	096	319	-,174	191	217	279	291	56	- 41	¥ġ¥
لستتسا					-303					1 -500	4						<u> </u>	<u> </u>	L-,	<u> </u>	

Orifice						ettac				
No.	-30	-10	℃.	10	20	†o	60	80	ង	120
67									1	
68	-0.063	-0.131	-0176	-016		-0555			-0900	
69	.009	056	309	126	138	147	173	226	3IA	33
70										i – – .
71										
72	343		- 208			.005	-105	.185	.266	.34
73	304	200	145	094	077	-022	.098	.174	.245	-30
74				.196	118		-002	.061	.107	16
15 76		192	172			064	044	-013	.051	10
70	218		174			197				-05
77	160	158				037	.014	-075		.10
300 IO	- 201	l∷i‰				090		017	.œ6	
79 86	- 166	175	12				068		016	.03
ã	- 224		028			448	763		728	73
ãe	043	00			516		- 586		720	
Ãa	003		1kg			436	772		728	
83 84	.001		103			~.327	199	509	622	
85	018		103				- 323	437	535	59
86	087	- 149	180		211	244	278	366	498	7
87	167	225	266	985	291	336	- 374	- · for	708	احا
87 88	155	204	219	236	27	28+	عنو	321	439	48
89	loð6	169	199	200	940	276	237	236	408	
90	42 6		113		032		.326	399	.446	1 .48
91	461.		226	160	089	.024		جيد.	.286	
92	39 \	29 4	218	144	091					
93	333		166		079			.172		
9	188	1177	158		103		 ∞3	-Oht		
95 96	242				196			060		
96	260	245						OH7		
97 98	223				203		106			
96	182	205	209	205	195	[INS	1IX	079	010	լ.∝



COMPANDEMINT AT

TABLE IV.- CONTINUED
(b) Basic model, M = 1.3

rifice					Angle o	of atte	e k				Orifice				Ā	arje og	attec	k			
Zo.	9	4	8	30	90	Ŷ0	60	80	700	150	No.	-30	-10	00	70	20	J40	60	180	108	11
٥	1.438	1.437	1.440	1.449	1.463	1.459	1.454	3.445	2.432	1.411	34	-0.104	-0.091	0.060	0.046	0.00	0.000	0.072	0.196	_	_
1]					I	l					344	08%	059	032	011	014				.917	
8	. 203	-10	-+33	.431	.389	.336	.291	.244	-203	.136	35 36	פננ	017	087	066	038	0	.049	.096	-159	. 4
3]	+30	.386	.356	.322 .85	:139	.262	.219	-176	140	.099	36	1	131	110	086	060			.056	وند.	
2	.662	-250	.526		22	397	.346	.296	-853	.809	37	226	215	206	182	160		08e	043	.001	
3 1	.299 .047	- 쪼	.520	.498	- 179	.440	-396	.351	-312	.279	36	337	~332	331	320	299	207	857	225	190	
: I		-021	008	027	038	063	098	124	122	173	39	.141	.070	-001	074	198					
ا ا	997	- 303	,325	318 065	- 333	- 329	386	- 397	362	369	AL .	-035	027	184	-그것	209			38		
8 1	.072		.077	.012	.008	019		003	065	- 091	42	.030.	008 006	059 056	107	229			49		
امُد	.064	.050	.015	.002	007	042		092	1::117	-:146		.007	023	054	080	101	147 140				1::
n l			-015				00		J_*'		12	060	089	-::::	-141	160		227		336 297	II:
<u> 10</u>	-157	.188	و12.	.246	.280	. 110	.039	.436	.497	.770	15	080	100	123	141			225		276	
	-044	029	aŭ	.ou	.036	-332 -069	.115	.156	.208	.256 I	46	048	073	103	129			906			
끊	044	029	013	.006	.024	049	.093	-132	.170	.eie	47		L-:•:3.	1	L		1		[::5		15-
15	.240	.235	.231	-232	.243	.243	.270	.298		.962	44	054	080	102	196	138	169	902	233	251	ļ
عُد	062	055	- 039	o <u>š</u> o	.00	,oro	.068	الحقد ا	130	.243	19	- 949	163	000	005	.076		837	335	.414	
17	-602	- 20	.463	.411	.367	.302	.239	.17.7	-075	-008	50		L						1		1-1
18	-076	.010	037	087	119	176	240	300	360	407	51		 -	l:			1		- <i>-</i> -	-	I
19											52	200	171	135	095	057	001	.066		.216	۱.
20	197	الكِل									粱	160	170	196	093		006	.055	٠.١١٨	.178	
20.	okg	087					246	286	331	387	, <u>*</u>	197	180	160	13		066		-032	.087	١.
22	.085	-070	.021	00	023			130		197	55 56	193	180	163	139	114	08%	037	-007	-052	
왔	.059 .054	.096		025	043			149		201	I 2°		ተ ኤ -	- :	1	1 :	ł	17.5	l		<u> </u>
===	.029				668				369	229	77	15k	161	-::温	140 163		101	068		.017	۰
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ži							= = =	[]]	= = =	I	20	.040	015	104	183		- 333			33	
26 I	269	280	296	- 300	317	340	359	379	394	419	61	.023	030	086	139		I::##	371			
29	305	- 261			141	034	.134	.181	-375	490	60	.00	028	OT5	123		290		45		
30	306	393		355		- 257	151	045	.107	.183	63	.ooi	036	076	100		165			- 443	<u> -:</u>
33.	139	199	094	070	053	030	.019	.038	-137	.144	64	-030	067	097	119	136	174	216	307	362	
32				029		.096	.065	.111	.180	.215	65	094	مُند.۔ ا	فاد	170		ai5				
33 I	109	089	065	01	oze	.024	.orè	.130	.209	.272	66	307	135	158	178	193	216	231	298	310	(·

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ı.	-30	-10	0	10	Q.	Į,	B	80	5	120
67					- 1			-		
66	-0.098	وعدها	-0.151	0.163	0.178	-0.207	0.236	-0.886	كاوها	-0315
69	-072	-000	111	-111	-146	-178	- 199	- 834	- 277	298
ΤÖ										
71		l		l	<u></u>	I			L	
72	289	-,250	198	-,150	_101	~.023	.068	.156	.233	.310
73	- 853	-917	-160	- 109	-068	.001	.075	156	.200	.266
73								_:	L	
70	902	-191	170	198	_110	067	011	.075	.30%	.166
75 76	-,930	-193	174	156	137	-100	047	019		
77	-206	- 201	186	172	1156	129	076	- 015	.025	068
描	-175	-33	137	197	_166	- 063	010	.035	.063	.100
Ť	- 207	_190		_16	140	- 100	-055	009	.027	.070
79 80	- 210	106	L 745	-167	_148	-115	-017	-039	005	.038
81.	.220	136	.030		- 143	- 872	376		- 326	576
82	.041	-031	100	186	- 895	- 111	- 10	161	- 511	- 500
81	-,008	o =-	-,117	194	-266	- 160	-410	504	510	565
83	.005	-033	083	-111	- 190	301	,391	- 478	523	550
85	-011	- 017	018	-11	- 145	- 237	331	-406	- 472	L.313
85 86	- 088	- 109	- 136	-177	-177	- 209		- 385	- 309	L 170
87	-,178	- 197	_ 991	- 0-0	- 257	263	- 110	- 343	- 186	- 46
87 86	177	- 193	907	213	- 203	- 244	-,273	- 290	320	-,398
89	-111	170	- 079	- 809	-22	_241	-243	-,134	661	⊢.359 l
90	-311	- 199	086	.019	.087	.186	.297	.367	.487	.479
91.	- 365	296	متع.ــا	قدا	09i	009	.090	.179	.249	.330
92	I588	351	- 258	192	_, 127	-033	.059	.144	فتعر	.330
93	- 297	- 258	-,202	139	09i	083	.060	24.3	.204	.275
93 94	- 234	917	186	_157	_110	076	024	.035	.063	.139
95	- 255	ندهــا	_ 222	203	104	-178	-114	058	019	.030
95 96	- 264	-815	- 225		188	162	114	- 023	.003	037
97	298	- 211	- 204	- 195	183	-,156	-110	-065	027	.020
á l		- 070	_ 000	L 166	_ 774	160	L 110	_ 07á	_ ^_	1

(c) Tests	TABLE I
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ough 38	

	88	13	8.	8	苏	9 3
	2842288 2016 pi	2747575 64 64 64 66	99%98989 99%98989	2988684 298664 2	8748427 66-66-66	Q
	.	4848636	- 101 - 102 - 103 - 103	हा देश के कि के कि इस इस इस के कि के कि	650 650 650 650 650 650 650 650 650 650	J.
	999999	\$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$	899950	599753	0.000 000 0.000 0.000 0.000 0.000 0.000 0.000 0.000 0.000 0.000 0.000 0.	કુ
	288888	200 A 6 8 8 8 8 8 8 8 8 8 8 8 8 8 8 8 8 8 8	882E88E	\$8283 \$833 \$8	98498E	ρ
		-0.0007 -0.0003 -0.0013 -0.0017	10018 10018 10018 10018 10018 10018	2000 2000 2000 2000 2000 2000 2000 200	.000) .000) .000)	c ₁
	26.54. 26.54.	. 455 455	18854H	#### 88958	88888	ī/p
34	<u> </u>	18	75	8	88	Teat
њ. В З	88888888 14. 44.00	9458928 6456	# 9 K 9 K K K K K K K K K K K K K K K K	: 4 . 94.00 25382558	9245 944 944 944 944 944 944	P
01.2 083	2 2 8 8 8 E 8	Federate	\$ \\\\\\\\\\\\\\\\\\\\\\\\\\\\\\\\\\\\	F95555	\$3555 \$3555	្ន
.0471 4740	9299 C C C C C C C C C C C C C C C C C C	\$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$	12000000000000000000000000000000000000	282228 2828 2828 2828	00000000000000000000000000000000000000	8
ig ig	998999	9893883B	3838388	323EFF8	2888888	S
,0011		0000 0000 0000 0000 0000 0000 0000 0000 0000	1 01100	9000 9000 9000 1100 1100 1100	00000 00000 00000 00000 00000 00000 0000	3
1 1	5.5.0 5.0		71588 T	8 R X F B : 1	発売的の からける。;;	둫
	**	37	<u> </u>		€ ¥	9 3
87.8	4 4 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5	4994d9P4	88888888888888888888888888888888888888	9884688	px	P
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167		2000 100 100 100 100 100 100 100 100 100	200000000000000000000000000000000000000		0.00 0.00 0.00 0.00 0.00 0.00 0.00 0.0	કુ
	38E 38E	8222228	286888888888888888888888888888888888888	111111	88 3365	P
.0016	0000		00000			c,
	8888 FFF	848438 848438	\$\$\$\$\$\$ \$\$\$\$\$		3883	둫

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88.087.9968888 86.087.9968888	87 78 F: 80:	8 8716878	25.00 5.1 5.1 6.2 5.0 5.0 5.0 5.0 5.0 5.0 5.0 5.0 5.0 5.0	1. % 6 1. % 6
565. 126. 126. 126. 126. 126. 126. 126. 126	£ 8 32.85	3 18 8 2 8 3 8 3 8 3 8 3 8 3 8 3 8 3 8 3 8	4 4 8 8 8 8 8 8 8 8 8 8 8 8 8 8 8 8 8 8	£8£8
	.017.0 .017.0 .017.0 .017.0	1 2 2 2 3 3 3 B		2200 XYE 6
\$0.50.50.50.50.50.50.50.50.50.50.50.50.50		2	3665 3665	9888
	0013 0013	00000 1000 1000 1000 1000 1000 1000 10	0015 0005 0017 0017 00201 00201	- 0012 - 00012
\$ \$ 2 \$ 3 k3 k	3.72	8 1 3 2 2	######################################	8.4.
	7	#	<u></u>	ίζ
<u>%&%%%%%%%%%%%</u> %pp%%*%*	1839E8 4	7. 1. 1. 1. 1. 1. 1. 1. 1. 1. 1. 1. 1. 1.	. 4.44 5.844.4885 7.444.8844.8855	17.23 19.29 21.33
266 266 266 266 266 266 266	BESES		2020 2020 2020 2020 2020 2020 2020 20	0.8g.
.0443 .0743 .0743 .0348 .0376 .0450 .0703 .0703 .1709	65 65 65 65 65 65 65 65 65 65 65 65 65 6		25.55.55.55.55.55.55.55.55.55.55.55.55.5	0.24.33 2004 3409
077	9984	20 10 10 10 10 10 10 10 10 10 10 10 10 10	000000000000000000000000000000000000000	986.
.0004 .0007 .0024 .0027 .0027 .0027	9000		100 100	0005 0005
86544865 6646866446				98.53 5.53 5.53 5.53 5.53 5.53 5.53 5.53
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\$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$	7.5° ₹ 888 7.7° † . 12° £	**************************************	**************************************	20,8 Å
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200 - 120 -		\$ 5 \$	004 075 075 152 075	0.00 0.00 0.00 0.00 0.00
0017 0017 0017 0017 0017	0008	94 5000- 94 5000- 97 1 1000- 97 1 1000- 98 9 9 9 9 9 9 9 9 9 9 9 9 9 9 9 9 9 9	0010 0.86 0011 1.30 0012 2:73 0005 1.26 0005 3.88 0013 3.77	1 1 2 2
. 1964.66	3.70	14446H	33777010	111

TABLE III.- CONCLUDED
(d) Tests 39 through 48



TABLE IV.- EXPERIMENTAL PRESSURE COEFFICIENTS, C_p (a) Basic model, M = 1.2

· ·	_											_									
Orifice				<u>An</u>	gle of	* attac	ik				Orifice						f atta				
No.	-30	4	00	10	δo	ħ0	8	9	ង្គ	120	Ec.	9	-10	တ	10	Q	9	B	8	100	9
0	1.354	1.377	1.386	1.368	1-399	1,384	1_392	上392	1.376	1.372	34	-0.103	-0.018	-0001			0.01	0.109	Ġ	0.222	68
1 1				- <i>-</i> -						f	344	115	07	062		003	.038	.101	.164	.223	.290
1 2	-89	.+38	-104	.981	.35	.296 .289	150	.27 .11	.153 .098	ميد.	35 36	126	104 117	085	053	032	.aia	-074	-138	.187	-237 -169
3	.409	-358	.326	337	-250	.229	-750	.147	.098	-067		1 1 1 226	#i	102	069 172	072 157	003 121	.046 081	-01	-135 007	.038
i <u>*</u>	.607	.596 .580	.581 .495	·23	:733	403	.300	35	·935	.28	H 50 50 50 50 50 50 50 50 50 50 50 50 50	708	- 35	- 373	340	- 333	30	277	251	227	185
1 3 1	-,082	1.25		1.470	23	146	.368 168	192	.286 .197	200	1 2 5	.179	.030	02		- 228	- 33	479	- 56	- 678	737
1 2 1	- 3-5	062 367	061 406	096 417	- 33	147	- 440		- 105	- 393	5 6	.057	- 02	068	172	243	- 318		770	656	706
1 4 1	.108	-104	095	.103	.095	.082	069	.67	.031	- 674	41	.016	032		- 126	175	391	446	55	624	675
	.002	034	019	668	œ	126	166	- 190	226		No.	.032	036	074	-:118	115	167	248	420	::23	6 12
امُدا	.059	aro.	003	016	035	070	094	عندا	147	179	報	016	016		09h	199	175	209	266		603
l ii								<u></u>	⊢		44	040	103	199		184	232	269	296	310	300
12	.109	.158	-181	.215	.248	.040	.361	.117	. 469		45 46	067	103	123		168	818	248	286	315	286
13	093	066	047	017	.002	.oto	.097	.145	.187	.270	₩ 20	030	085	107	126	172	196	223	259	279	263
14-	072	055	034	.015 .245	.002	.035 .036	.097 .082 .260	-123	그것	1.205	47 48	-029	081	113	194	139	166	204	240	- 275	198
15 16	.232	.239	-237	-247	200	-22	.260	.270	.267	.264 .251	19					.086	.191	.984	-572	潂	- 300
15	.160 316	056 056	037 355	008 .299	.160	377	.119		-937 093	1.33	- 36						1				
協	.008	083	136	33	222	301	561	-003	- 22		カルスの大		- <i>-</i> -		l <u>-</u>	i	l		- <i>-</i> -		
19		1003	[22	222	161	118	072	010		-108	.182	.247	.32*
ãó	114	178	200	239	265	332	368	432	37.4	27	- 53	160	137	6بد		054	.012	.079	.IA6	.209 .094 .041	-279
<u> </u>	.024	006	.017	061	085	131	167	207	261	366	> .	209	186			118		.003	.076	.09*	75
22	.094	.okk	.022	005	018	073	115	159	197	226	72	178	175	15%	183	108	o c a	029	.009	.04⊒	-085
23	.069	.026] .001	022	070	094	126		199		7×7×76		تـــ تا	1- :		l	<u>-م</u> ـ	015	I- ==		-075
24	-053	-010	017	038	078	096	130	162	805	229	길	- 16	126 177	136 168		111	063	071	010	.051 .005	.050
85	.077	007	037	060	08IL	128	159	193	235	271	L 26	.185	0.05	026			369	522	- 651	- 716	747
86			{	- <i>-</i> -			r	r	<i>-</i>	r	26	017	06	111	195		436	三菱	611	639	733
27 26	264	- <u></u> -	325		344	366	323	F. 845	.20	173	ã	.027	068	119	-174	226	398	157	606	- 619	723
29	- kok	달	1.32	330	222	085	Г:ॐ	T:564	T.305		62	.016	032	092	- 343		228	107	54	695	- 700
36	- 473	.66	- <u>2</u>		309	- 23	.086		1.35		63 64	004	069		1kī	J1¥8	209	517	463	563	672
ı	- 146	115	110	002	068	044	.005	.036	.007	.26 .12	64	033	098	126	143	175	215	J064	- 404	535	624
1 52	097	067	046	020	.005	.041	.087	139	.194	.291	65	018	138	172	197	221	263	290	<i>3</i> 72		- 770
39	111		065	.060	009	.041	פני.	.178	.233		66	096	3/19	174	191	 217	279	991	56	- Hi	191
لسئتسا						<u> </u>					4						_				

Orifice						ettac				
No.	-30	-10	℃.	10	20	†o	60	80	ង	120
67									1	
68	-0.063	-0.131	-0176	-016		-0555			-0900	
69	.009	056	309	126	138	147	173	226	3IA	33
70										i – – .
71										
72	343		- 208			.005		.185	.266	.34
73	304	200	145	094	077	-022	-098	.174	.245	-30
74				.196	118		-002	.061	.107	16
15 76		192	172			064		-013	.051	10
70	218		174			197				-05
77	160	158				037		-075		.10
300 IO	- 201	l∷i‰				090		017	.œ6	
79 86	- 166	175	12				068		016	.03
æ	- 224		028			448	561		728	73
ãe	043	00			516		- 300		720	
Ãg	003		1kg			436			728	
83 84	.001		103			~.327	199	509	622	
85	018		103				- 323	437	535	59
86	087	- 149	180		211	244	278	366	498	7
87	167	225	266	985	291	336	- 374	- · for	708	اج
87 88	155	204	219	236	27	28+	عنو.۔	321	439	48
89	loð6	169	199	200	940	276	237	236	408	
90	42 6		113		032		.326	399	.446	1 .48
91.	461.		226	160	089	.024		جيد.	.286	
92	39 \	294	218	144	091					
93	333		166		079			.172		
9	188	1177	158		103			-Oht		
95 96	242				196			060		
96	260	245						OH7		
97 98	223				203		106			
96	182	205	209	205	195	[INS	1 <u>1</u> }	079	010] -oc



COMPANDEMINT AT

TABLE IV.- CONTINUED
(b) Basic model, M = 1.3

rifice					ingle o	of atte	e k				Orifice				Ā	agle of	attec	k			
Zo.	9	4	8	30	8	Ŷ.	60	80	700	150	No.	-30	-10	00	70	20	J40	60	180	108	11
٥	1.438	1.437	1.440	1.449	1.463	1.459	1.454	3.445	2.432	1.411	34	-0.10	-0.091	0.060	0.046	0.00	0.000	0.072	0.196	_	_
1]						l					34A	084	059	032	011	014				.917	
8	. 203	-10	-+33	.431	.389	.336	.291	.244	-203	.136	35 36	פננ	017	087	066	038	0	.049	.096	-159	. 4
3]	+30	.386	.356	.322 .85	139	.262	.219	-176	140	.099	36	1	131	110	088	060			.056	وند.	
2	.662	-250	.526		- 29	397	.346	.296	-853	.809	37	226	215	206	182	160		08e	043	.001	
3 1	.299 .047	- 쪼	.520	.498	- 479	.440	-396	.351	-312	-279	36	357	~332	331	320	299	207	857	225	190	
. I		-021	008	027	038	063	098	124	122	172	39	.141	.070	-001	074	126			403		
ا ا	997	- 303	,225	318 065	- 353	- 329	386	- 397	362		A1	.035	027	184		209			38		
8 1	.072		.077	.012	.008	019		003	065	- 091	40	.037	008 006	059 056	107	222			49		
امُد	.064	.050	.015	.002	007	042		092	1::117	146		.007	023	054	080	101	147 140				1::
n l			-015		00		00		J_*'		12	060	089	-::::	-141	160		227		336 297	II:
<u> 10</u>	-157	.188	و12.	.246	-260	. 110	.039	.436	.497	.550	45	080	100	123	141			225		276	
	-044	029	aŭ	.ou	.030	-332 -069	.115	.156	.208	.256	46	048	073	103	- 129			906			
끊	044	029	013	.006	.024	019	.093	-132	.170	.eie	47		L-:•:3.	1	L		1		[::5		15-
15	.240	.235	.231	-232	.243	.243	.270	.298		.960	440	05%	080	102	196	138	169	902	233	251	ļ
عُد	062	055	- 039	o <u>š</u> o	.00	,oro	.068	الحقد ا	130	.243	19	- 949	163	000	005	.076		837	335	.414	
17	-602	- 20	.463	.411	.367	.302	.239	.17.7	-075	-008	50		L						1		1-1
18	-076	.010	037	087	119	176	240	300	360	407	51	l	 -	l:			1		- <i>-</i> -	-	I
19										- - -	52	200	171	135	095	057	001	.066		.216	۱.
20	197	الكِل			249						翠	160	170	196	093		006	.055	٠.١١٨	.178	
20.	okg	087					246	286	331	387	*	197	180	160	13		066		-032	.087	١.
22	.085	-070	.021	00	023			130		197	55 56	193	180	163	139	114	08%	037	-007	-052	
왔	.059 .054	.096		025	043			149		201			ta.	- :		-:-	ł	17.5	l		<u> </u>
===	.029				068				369		7	15k	161	-::温	140 163		101	068		.017	۰
න නේ		Ľi	033	025			130					.198	193 -097	-:010	071	139 139		343		- 014	
ži							= = =	[]]	= = =		26	.040	015	104	183		- 333			33	
26 I	269	280	296	- 300	317	340	359	379	394	419	61	.023	030	086	139		I::##	371			
29	305	- 261			141	034	.134	.181	-375	.490	60	.00	028	OT5	- 183		290		45		
30	306	393		355		- 277	151	045	.107	.183	63	.ooi	036	076	100		165			- 443	<u> -:</u>
33.	139	199	094	070	053	030	.019	.038	-137	.144	64	-030	067	097	119	136	174	216	307	362	
32				029		.096	.065	.111	.180	.215	65	094	مُند.۔ ا	فاد	170		ai5				
33 I	109	089	065	01	012	.024	.ore	.130	.209	.272	66	307	135	158	178	- 193	216	231	298	310	[·

Orlfice						of at	tack			
Ho.	-30	-10	0	10	Q.	Į,	B	80_	5	120
67					- 1					
66	-0.098	وعدها	-0.151	0.163	0.178	-0.207	-0.236	-0.886	كاوها	-0315
69	-072	-000	111	-111	-146	-178	- 199	234	- 277	298
ΤÖ										
71		 		 					L	
72	289	-,250	198	-,150	_101	~.023	.068	.156	.833	.310
75	- 953	-917	-160	- 109	-068	.001	.075	156	.200	.266
73										
73	902	-191	170	198	_118	067	-011	.077	.30%	.166
75 76	-,210	-193	L174	156	137	-100	047	019	.066	190
77	-206	- 201	186	172	1156	129	076	-015	.025	068
描	-175	-11	-137	197	- 100	063	018	.035	.063	.100
79	- 207	_190		16	140	- 100	-,055	009	.027	.070
79 80	210	_106	_ 1A4	_167	_148	-115	-017	039	005	.030
81.	.220	136	.030		- 143	- 872	376	I 1677.	- 326	576
82	.041	- 031	100	189	- 855	341	- 10.	- 181	-,541	582
83	-,000	-054	-,117	190	-265	360	-439	504	550	565
8.	.005	-033	083	-111	- 190	301	,391	-3-78	323	550
85	-011	- 017	018	-11	-145	- 237	331	-,406	-,472	L.313
85 86	- 088	- 109	- 136	-177	-,177	- 209	560	-385	- 309	اقطاسا
87	-,178	- 197	_ 221	- 9-8	- 257	203	318	-343	- 386	- 46
87 86	- 177	- 193		213	- 200	844	-,273	- 290	328	-,398
89	- 139	170	-079	- 209	_22	_241	-243	-,13k	661	⊢.359 l
90	-311	- 199	086	.019	.087	.186	.997	.367	.407	.479
91.	- 365	296	متع.ـــا	قدا	091	009	.090	.179	.949	1 .330
92	588	351	- 258	192	_, 127	-033	.059	.144	<u> مدع</u>	.330
93	297	- 258	-,202	139	091	023	.060	.243	.204	.215
93 94	- 234	917	186	_157	_110	076	094	.035	.063	.139
95	- 255	-81	_ 222	203	104	158	-114	058	019	.030
95 96	- 264	-815	- 225	- 207	-,188	160	114	055	.003	.037
97	298	- 211	- 204	- 195	183	-,156	متت	-065	027	.020
á l		- 070	_ 000	L 166		160	1110	L ~~	_ ^_	



TABLE IV.- CONTINUED (c) Basic model, M = 1.7

Orfice				Angle	of at	tteck					Orfice				Angl	of a	ttack				
No	-30	-10	တ	10	20	40	6°	80	100	120	Χo	-3°	-10	60	10	[2º	40	60	80	10	120
0	1.561	1.587	1.598	1.571	1.595	1.601	1.588	1.575	1-557	1.543	35	-0.063	-0,056	-0,034	-0012	0.006	0.046	0-097	0.136	0.176	0.224
1 1				- <u>-</u> -						l	36	081				015		.075	פנו.	.152	.2C#
2	-507	+57	+39	-405	.376		-291	-246	.207	.167	37	136	127	115				006	.029	.071	.114
3	. 19	-379	-356	.324	-302	.264	.224	-188	.152	-112	38	198	197	191		-165			095	067	035
:	.489 .631	-436	.427	.380	.356	.322 .466	.277	243	.203	.167 .261	39 40	.178		-090		-016			147	202	252
3	.190	.581 .156	.561 144	.521	.500 .097	.079	.010	.365 .010		- 011	1 1	011	011	032							297
7	124	150	162	175	189	197	219	230	245	249	42	.023			120			227	266		337
افا	080	065	099	113	131	103	051	025	00á	.006	43	.oiz		001		143		198			- 316
اوا	.053	.026		044	.046	049		.053	.035	.023	144	026		067		090		137		234	
10 1	.055	-030	.037	.033	.024	.016		026	058	087	45	026								186	
1 11											46	026	051	064	079	088	113	134	152	173	219
12	.207	•380	.263	-269	.307	.361	.414	.520	.522	.576	47					ļ					
13	.013	.026	-047	.068	.076	.116		.196	.240	.2 83	¥8	018					114				192
[14	005	-007	.020	.038	.050	.079	.118	.149	.187	.226	49	060	034	-004	.059	.098	-195	.295	.366	.441	-513
15	.195	-179	.176	.166	.161	.165	-169	.168	-175	.188	20										
16	019	016	005	.020	.029	.063	.112	.148	.194	.222	51 52	150	114	074	036	018	.035	.103	3 = 7	.219	.263
l iš l	.725	.661 .144	.638	.599 .101	·295	.512 .046	.407 041	.389 075	103	149	53	083			037			.097	.157 .137	.185	.244
19	- 190		-120	.101	000	.040		0/5			54	~.127			072		013	043	.082	129	.179
ão l	073	113	126	148	172	196	219	241	266	268		123					021	.02	.065	.109	.153
21	036	071	085			158				273	55 56] -									
22	.069	.037	.030	.016	-004	019	045			148	57	049			059			.006	.011	.080	.128
23	.053	.021		006		039		088		131	58	123		109				.003	.040	.075	.121
24	-054	.024		003		034		078		124	29	.221	-157	-126				091	161	221	264
25 26	-045	.008	-001	00.5	029	043	069	092	115	140	60	.060	057	018 086		088		184	- 230	276	312
20 27				- - -							62	.005				12		- 235	259	293	321
26 l	123	146	151	161	176	- 185	203	217	234	254	63	.020	040						276		- 341
a9	037	.002	.048	.124	260		.318	379	-550	.647	64	019	020	041				- 22	267		- 326
30	128		129		076	- 067	.055	:111	.234	-32	65	035		075				- 225			- 118
31	171	156		085	060	013	.042	.085	.122	166	66	052		094				217			
32	046	039			-004	.033	.075	.109	.150	.196	67										
33	048	032	019	.009	.025	.061	.112	.154	.197	.248	68	057					169				
34	056	041		008	.005	-05h	.099	.140	.180	.243	69	024	054	080	097	113	135	[1 <i>5</i> 8	176	205	232
344	071	054	045	005	*007	.048	-097	.141	-181	-235	70 .				i			- - -			
							\vdash					Ц	Ь			Ь—	Ь—			NAC	

Orfice				Angle	of a	ttack		_		
Eo Eo	-3°	-10	00	10	20	10	60	80	10°	12°
	-3_		Ŭ	_	_	-	Ť	_		
71					<u> </u>				2 2.5	,-
72		-0-158		-0011		0.023				
73	174	241	108	051	.020	.045	.111	.171	.236	.304
74							<u>-</u> -			
75	157			o <u>7</u> 3	055				.144	.199
76	128	124		085						.166
77	141			109			003			.140
78	109			077		017	.036			
79	142	140	I37	110	098	070	021	.018		
80	149			111			027	.008		
81	.278		-160	.119		021			271	308
82	.074		oi3	047	082	135	196	241	285	330
83	.010			098	132	175	I43	274	313	
84	0	055	.091	111	150	193	150	260		352
85	.017		092	118	148	187	142	258		325
86	.005		083	133	174	220	162	287		347
87	065			139	184	23I		313	340	364
88.	089	110	123	135	173	227	170		334	
89	091			129	12	209	163	289	290	301
90	042		.058	-I37	.185	.263			476	.532
91	136	1111	068	014	•020	.089				
92	184	146		070	043			·173		
93	185	(I63			039				-235	
94	141			081						
95	187		151	112	096	068	022			
96	193			118						
97	185	164	155	130	119	091	033		.036	
98	174	163	156	132	[120	094]0 % 1	019	.Out	.067



NACA RM A52J31

TABLE IV.- CONTINUED
(d) Model with rocket packets, M = 1.2

A1-A1 -		Angle	of at	tack				Angle	of att	ack			<u></u>	Angle	of at	tack	
Orifice No.	-3°	00	ļю	80	120	Orifice No.	-30	00	70	80	12º	Orifice No.	-3°	00	ηо	80	12º
0	1.360	1.375	1.384	1.371	1.384	34						67					
1						34A	014	.082	.176	.299	.363	68	083	166	246	365	50
2	.485	•395	.287	-193	.107	35	103	048	.042	.133	.194		- 055		207	316	
3	.401	.31.0	.218	.132	.057	36	230	164	042	•033	.168						
4	.608	.569	•436	.307	.212	37	248	175	095	019	.032	71					
5	•552	•477	•398	.320	.253	38	356	333	276	220	198	72	368	255	046	.170	.24
6	034	094	154	204	250	39	.159	123	497	704			347		128	.050	-33
7	379	 426	441	.408	418	40	.012	189	506	- 690							
8	.106	.091	•081	.051	•055	41	•033	122	430	644	702	75	210	121	008	.091	.14
9						42	.030	094	212	560	673	76	201	133	037	.042	10
10	.049	013	080	126	174	43	016	112	222	382	646	77	163			.004	.06
11) į .jį	064	155	25 5	310	333	78	173	1 -		.043	.08
12	.114	•193	.309	.424	•530	45	085	151	227	289	305	79	~.185		079	012	.06
13	060	036	.060	.157	.245	46	059	134	213	267	306	8o	134		113	044	.10
14	073	030	.047	-134	.201	47						81.		146		726	72
15	.226	.234	.256	•263	.271	48	018	138	194		287	82	.003	252	576		
16	.003	•059	.174	.271	•308	49	117	.065	.268	.384	-517	83	038	220		692	70
17	.505	.328	.153	031	162	50						84	025		458	619	68
18	.013	153	345	474	575	51						85	040		304	527	64
19						52	307	209	058	.164	.299	86	110	- 210	296	- 445	57
20	141	250	381	496	- 593	<i>5</i> 3	367	243	020	.116	.167	87		301	373	- 445	
21	-034	060	135	-,227	397	54	177	108	0	.085	.158	88		- 244	- 297	337	51
22	.082	.007	080	167	247	<i>5</i> 5	173	126	049	.015	.071	89	109		244	- 276	
23	.058	016	108	182	256	56						90	297		.212	387	.47
24	.044	034	108	183	250	<i>5</i> 7	157	103		.033	.050	91		166		.186	-35
25	.051	035	125	203	265	58	185	157	083	005	.052	92	365		049	.203	31
26						59	.129	168	610	753	753	93	351		.007	.206	.27
27						60	.002	246	577	715	752	94	251		.056	.135	.16
28	305	337	336	220	239	61	019	210	558	701	729	95	251	201	123	041	.01
29	400	285	045	-183	.408	62	027	143	287	653	708	96		188	131	051	.00
30	394	270	063	.133	.217	63	030	139	272	581	668	97		169	129	050	.02
31.	155	087	012	.096	.244	64	061	157	271	518	642	98	147	166		067	.00
32	224	.244	.290	-326	-357	65	105	203	313				"	i		''	
33	343	285	134	.039	.087	66	131	197	301	417	560			l	1		

TABLE IV.- CONTINUED

(e) Model with rocket packets, M = 1.3

		Angle	of at	tack				Angle	of at	tack				Angle	of a	ttack	
Orifice No.	-3°	00	140	80	120	Orifice No.	-30	00	μо	80	12 ⁰	Orifice No.	-3°	00	цо	80	12 ⁰
0	1.417	1.442	1.440	1.428	1.407	34						67					
1						3 ¹ tA	.047	.012	.124	.245	-395	68	~.099	139	210	262	414
2	.508	.43I	.321	.228	.151	35	093	046	.041	.134	.236	69	060	115	175	226	315
3	.419	.346	.247	.163	.083	36	209	162	~.075	.026	.135	70					~~~~
4	.640	.522	-377	.278	.205	37	258	220	128	030	.047	71.					
5	.585	518	.417	-337	.266	38	320	~.309	269	222	156	72	310	226	092	.146	.268
6	.040	012	053	128		39	.128	023	255	434	551	73	325	233	084	.003	.209
7	293	334	378	396		40	.043	105	- 303	453	- 550	74		*****	****		~~~~
8	020	•057	.065	.081	.055	41.	011	061	- 267	1414	544	75	226	14I	~-040	.082	.178
9						42	008	054	154	- 355	507	76	210	155	- 065	.038	125
10	.054	,009	050	101	- 155	43	.003	~.065	155	296	431	77	209	171	084	.015	.089
n i						74	068	127	-,204	273	381	78	176		~.043	051	123
12	.143	226	.325	.445	.561	45	084	126		265	277	<u> </u> 79	195	156	075	003	.054
13	055	004	.066		.265	46	050	107	182	235	2 5 6	80	194	162	092		.024
14	055	011	.052	.132	.225	47						8T	.180	034	285	-:453	- 565
15	.220	.231	.232	252	.267	48	062	106	172	238	- 268	82	024	136	3 <u>48</u>		- 567
16	046	009	.085	.231	.348	49	226	-023	.209	.358	.446	83	016	1 41	358		 578
17	-580	450	.288	.125	027	50			P		****	84	•00/	098	- 315		- 562
18	.060	~.054	- 192	315	419	51.						85	012	086	246		524
19				HAHE		52	274	172	-,025	.085	.292	86	085	138	208		464
20	- 139	218			487	53	324	- 299	170	.053	.183	87	174	228 -	282		- 464
21.	027	089			397	外	175	115	036	.072	.170	88	173	209	229		415
22	071	.018			202	55	177	136	064	.021	.095	89	140	j191	223	- 238	390
23	054	.005			~.209	56						90	227	045	-174	357	.465
24	.041	005			207	57	126		~.087	015	.061	91	311	~.192	013	15첫	-300
25	.026	020	096	-,163	219	58	173	- 159	095	017	.062	92	321	218	062	.067	325
26						59	.169	050			590	93	307	236	087	.138	.304
27						60	.025	151	369	汉;	592	94	267	229	086		193
26	260	296		- 408	412	61.	.006	114	316	468	561	95	284	212	-,136		.032
29	313	227	019	.214	£44.	62	.008	093		462	559	96	266	208	135		.032
30	- 345	353		036	.190	63	008	093	180	418	- 528	97	235	F-190	131	- 045	.031
31	235	089	-	.062	.171	64	042	102	178	348	467	98	21.6	h.189	134	062	.006
32	.230	.283		.361	-397	65	095		218		457	1	l	1			I
33	- . 348	~•3 03	207	052	.134	66	114	158	228	301	418		l			1	I

TABLE IV.- CONCLUDED

(f) Model with rocket packets, M = 1.7

	Ĺ	Angle	of at	teck				Angle	of at	tack		Γ'		Angle	of at	took	
Orifice No.	-3°	00	μo	80	12 ⁰	Orifice No.	-3°	0°	40	8º	12 ⁰	Orifice	-3°	00	40	80	120
No. 0 1 2 3 4 5 6 7 8 9 10 11 2 3 14 5 16 7 8 9 20 11 2 3 14 5 16 7 8 9 20 21 22 22 22 22 22 22 22 22 22 22 22 22	1.573 .506 .420 .489 .633 .195 -117 062 .056 .017 .034 .193 040 .723 .199 026 .075 .053 .053	1.584 .433 .351 .413 .556 .141 .152 .105 .047 .026 .636 .128 .146 .105 .035 .005 .008 .148 .005 .005 .008	1.566 .338 .263 .314 .554 .087 .186 .186 .197 .197 .197 .198 .199 .199	1.578 .255 .190 .241 .361 .030 .033 .467 .115 .391 .046 .289 .289 .289 .289		10. 334 4556 78 394 444 44 454 78 455 555 558 566 866	141 026 .046 039 171 183 .176 .036 014 023 020 023 024 029 024 029 020 021 029 021 029 021 029 021 029 021 029 021 029 021 029 021 029 020 0	150 018 029 021 166 032 059 073 064 053 054 053 054 052 052 054 052 054 052 055 054 056 -	15 15 15 15 15 15 15 15	- 645 - 1666 - 1673 - 1	.056 .249 .271 .260 .233 .244 .245 .245 .245 .245 .245 .245 .245	#0 656 678 69 70 72 73 74 75 75 77 78 79 85 85 85 85 85 85 85 85 85 85 85 85 85	033 048 063 07 146 1178 1178 1174 136 137 138 136 088 088 088 088 166 166	081 084 151 123 123 024 024	1681751861210590050150160780161641811206207198122020066078020102	238 221 219 163 072 .074 .033 033 033 220 247 222 232 232	276 265 203 275 .238 193 .141



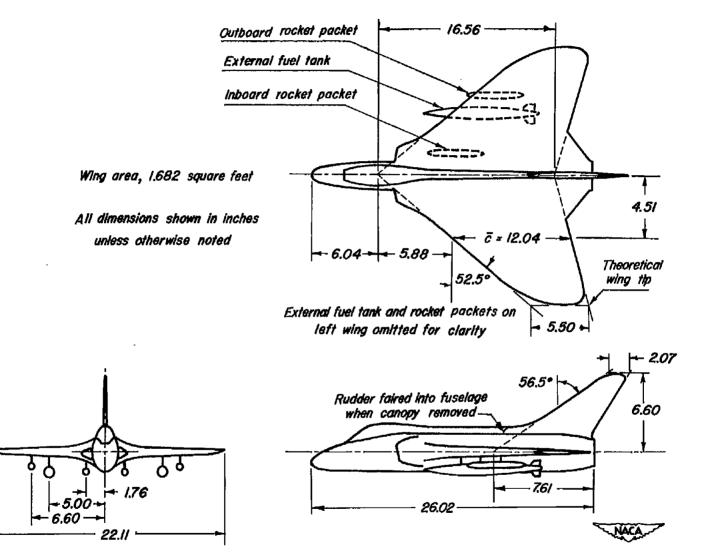


Figure 1.- Three-view drawing of the model showing the external fuel tanks and rocket packets.

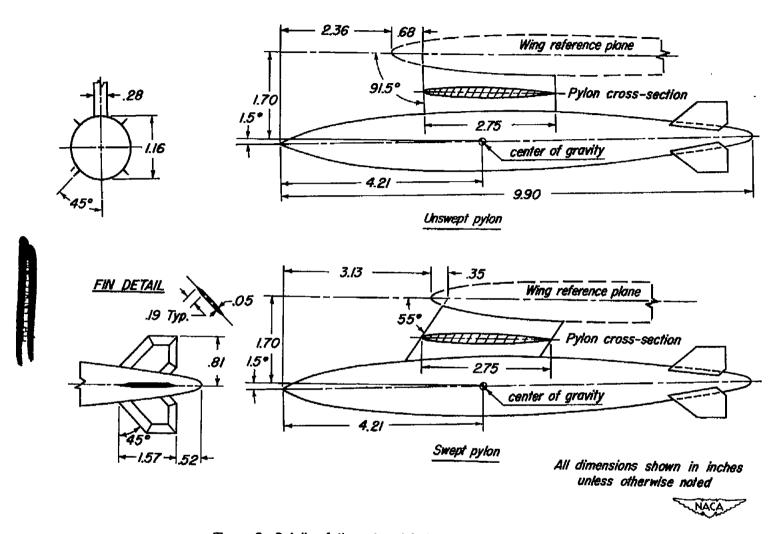
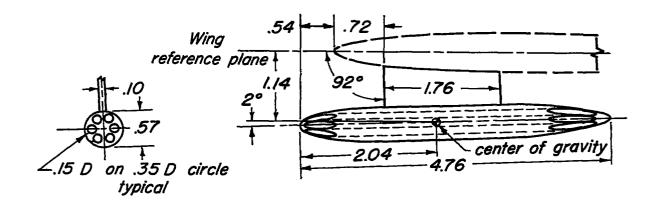


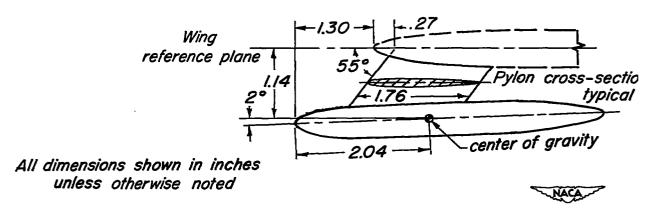
Figure 2.- Details of the external fuel tanks with unswept and swept pylons.





Note: rocket packet shown with open tubes

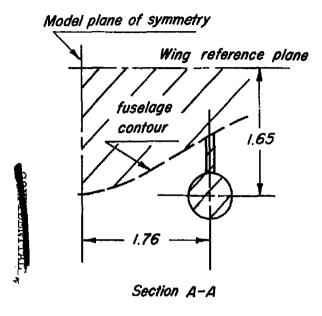
Unswept pylon



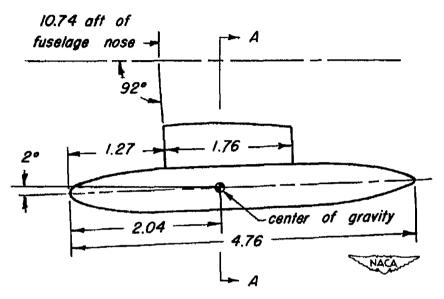
Swept pylon

(a) Outboard location.

Figure 3.- Details of the rocket packets with unswept and swept pylons.



All dimensions shown in inches unless otherwise noted



(b) Inboard location.

Figure 3. - Concluded.

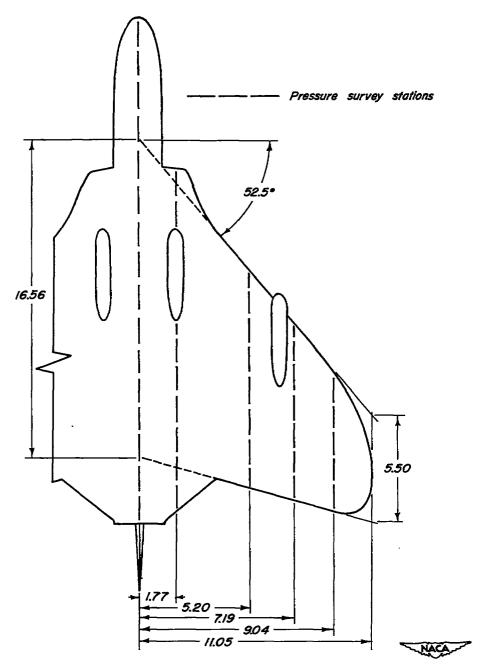


Figure 4. – Dimension sketch of the lower surface of the model with rocket packets installed, showing the pressure survey station.

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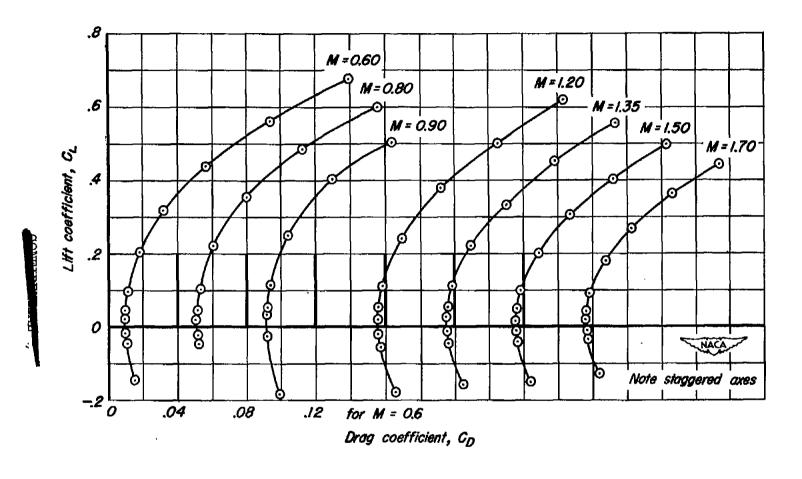


Figure 5.- Variation of drag coefficient with lift coefficient for the basic model.

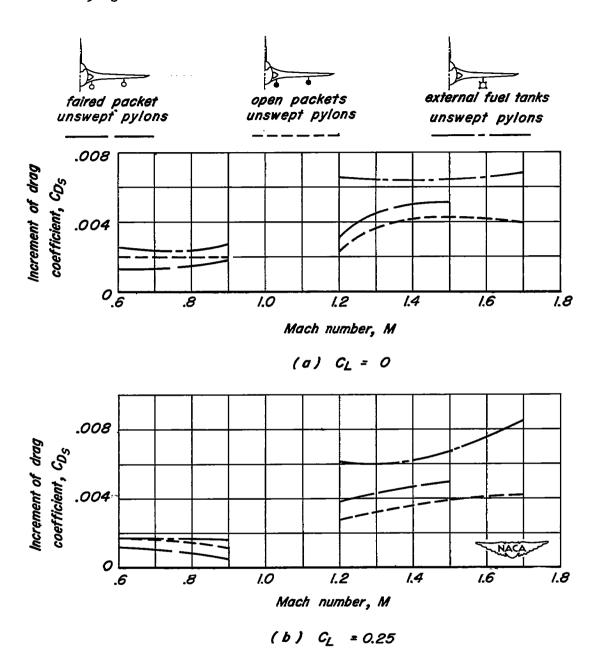
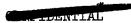


Figure 6.-Variation of Increment of drag coefficient with Mach number at O and O.25 lift coefficient for the various external store configurations mounted on the model.



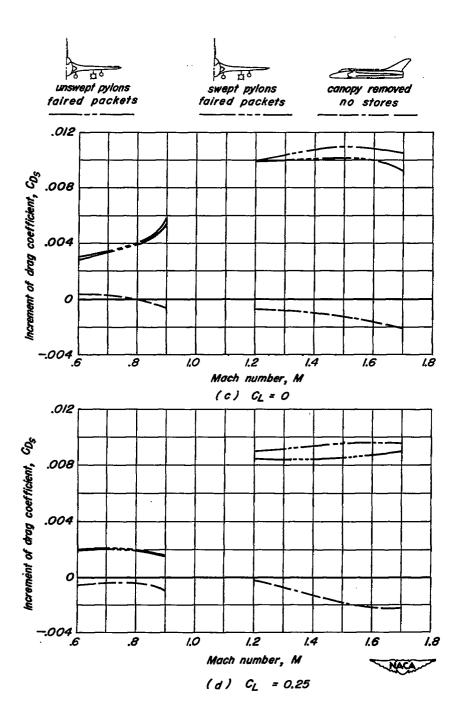


Figure 6 .- Concluded.



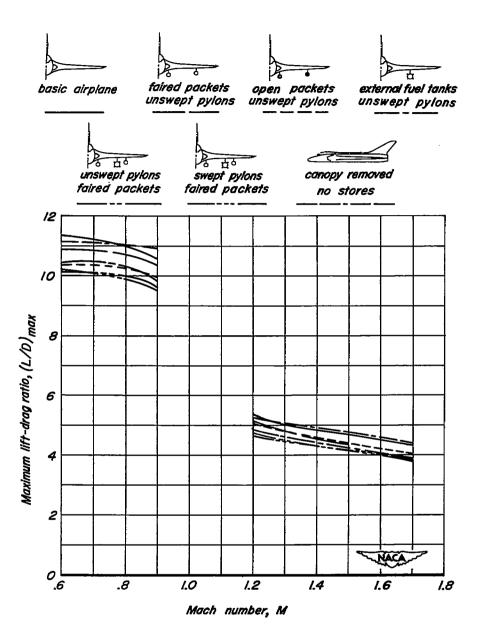


Figure 7.- Variation of the maximum lift-drag ratio with Mach number for the various external store configurations mounted on the model.

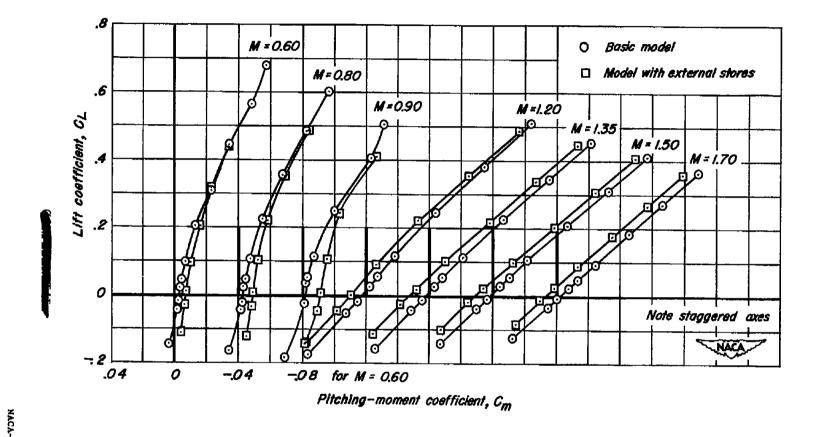


Figure 8.—Variation of pitching-moment coefficient with lift coefficient for the basic model and for the model fitted with two external fuel tanks and four faired rocket packets mounted on unswept pylons.

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